

IN THE CLAIMS:

This listing of claims below will replace all prior versions, and listings, of claims in the application:

Listing of Claims

Claims 1 to 16 (cancelled).

Claims 17 to 25 (cancelled).

Claim 26 (original): A method for milling recesses into a workpiece, comprising:

providing a milling tool, the milling tool including a base body and at least one cutting body situated on an outer periphery of the base body, the cutting body being angled in relation to the base body, the base body being disk shaped or plate shaped; and
milling recesses into a workpiece with the milling tool.

Claim 27 (original): The method as recited in Claim 26, wherein during milling, a rotation axis of the milling tool and a surface of the workpiece into which a circular recess is milled, enclose an angle which is greater than 0° and smaller than 90°.

Claim 28 (original): The method as recited in Claim 26, wherein the angle corresponds approximately to a second angle between an outside surface of the cutting body and a disk-shaped or plate-shaped surface of the base body.

Claim 29 (original): The method as recited in Claim 26, further comprising:

calculating a suitable milling radius and a suitable angle between a outside of the cutting body of the milling tool and the disk-shaped or plate-shaped surface of the base body of the milling tool based on: a radius, a depth and a width of the circular recess to be milled; and a permissible tolerance for the recess.

Claim 30 (original): The method as recited in Claim 29, wherein the tolerance includes a

tolerance for a circular inner wall and/or a tolerance for a circular outer wall of the circular recess to be milled.

Claim 31 (original): The method of claim 26, wherein the work piece is a gas turbine component.

Claim 32 (original): The method of claim 26, wherein the milling step comprises reconditioning groove-shaped recesses on gas turbine components which are deformed.

Claim 33 (original): The method of claim 26, wherein the milling step comprises milling flow channels between adjacent blades or for milling blade clearances during the manufacture of integrally bladed rotors of a gas turbine.

Claim 34 (original): The method of claim 26, wherein the milling step comprises milling single-blade profiles for gas turbine blades.